

Assessment of Stress-blended Eddy Simulation Model for Accurate Performance Prediction of Vertical Axis Wind Turbine

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Abstract

Purpose

The aim of this paper is to assess the ability of a stress-blended eddy simulation (SBES) turbulence model to predict the performance of a three-straight-bladed vertical axis wind turbine (VAWT). The grid sensitivity study is carried out to evaluate the simulation accuracy.

Design/methodology/approach

The unsteady Reynolds-averaged Navier-Stokes (URANS) equations are solved by Computational Fluid Dynamics (CFD) technique. Two types of grid topology around the blades, namely O-grid (OG) and C-grid (CG) types are considered for grid sensitivity studies.

Findings

Simulation results have shown significant improvements of predictions by employing the SBES turbulence model than that of other turbulence models such as $k-\varepsilon$ model with regard to the power coefficient (C_p). The C_p distributions predicted by applying the CG mesh are in better agreement with the experimental data than that by the OG mesh.

Research limitations/implications

The current study provides some new insights of the use of SBES turbulence model in VAWT CFD simulation.

Practical implications

The SBES turbulence model can significantly improve the numerical accuracy on predicting the VAWT performance at lower tip speed ratio (TSR) for which other turbulence models cannot achieve. Furthermore, it has less computational demand for the finer grid resolution used in the RANS-Large Eddy Simulation (LES) “transition” zone, compared to other Hybrid RANS-LES models.

Originality/value

To authors’ knowledge, this is the first attempt to apply SBES turbulence model to predict VAWT performance resulting in accurate CFD results. The better prediction can increase the credibility of computational evaluation of a new or an improving configuration of VAWT.

Keywords

Vertical axis wind turbine; stress-blended eddy simulation; grid topology sensitivity.

Nomenclatures

A	: area of the simulated model (m^2)
A_s	: rotor swept area (m^2)
c	: chord length (mm)
C_m	: moment coefficient
C_p	: power coefficient
C_μ	: coefficient of turbulence viscosity formulation
D_{rotor}	: rotor diameter (mm)
H_{rotor}	: rotor height (mm)
H_{wind}	: wind tunnel height (mm)
k	: turbulence kinetic energy (m^2/s^2)
M	: moment force (N)
R	: turbine radius (m), i.e. from the blade pressure centre to the supporting rod centre
U_∞	: wind speed (m/s)
W_{wind}	: wind tunnel wide (mm)

Greek symbols

ε	: turbulence energy dissipation rate of k - ε based turbulence model (m^2/s^3)
σ	: solidity
τ	: stress tensor (N/m^2)
ν	: kinematic viscosity (m^2/s)
ω	: turbulence energy dissipation rate of k - ω based turbulence model (m^2/s^3), rate of turbine rotation (rad/s)
ρ	: fluid density (kg/m^3)

Subscripts

i, j	: grid position in x and y coordinates respectively
t	: turbulence

1. Introduction

High-cost of power grid installation and connection have led to the growing interests in developing effective and affordable wind turbines in urban areas. While **Horizontal Axis Wind Turbines** (HAWTs) are still dominant in wind power industry, they gradually become less competitive in urban areas than those **Vertical Axis Wind Turbines** (VAWTs). Compared to HAWTs, VAWTs benefit from lower installation and maintenance costs (Ghasemian, et al., 2017). Furthermore, as VAWT can operate with any wind directions, they do not need a dedicated yaw mechanism. This can largely increase the operational reliability and thus is more suitable for urban environment where multidirectional wind flow exists (Sutherland, et al., 2012). In addition, their resilient characteristics to the wake effect of upstream blades and the vibrant background turbulence have made them preserving the ability to generate more power than HAWTs in urban areas (Dabiri, 2011).

While the performance of wind turbine can be measured by full-scale model for on-site tests or reduced-scale model for wind tunnel experiments, there are growing trends to use **Computational Fluid Dynamics** (CFD) method to predict the turbine performance such as the power generation. One major challenge in CFD simulation of VAWTs is to model turbulent flow around the rotating blades and the wake-turbulence interactions. The majority of turbulence models are able to capture the time-averaged mean flow properties with steady Reynolds-averaged Navier-Stokes (RANS) approach, while the large-scale flow unsteadiness can be reproduced using **unsteady RANS** (URANS) simulations. Both are sufficient for most engineering applications. However, the RANS model could not capture the small-scale turbulence fluctuations, which are important for understanding the underlying flow physics. This could affect the accuracy of CFD predictions for VAWT performance. Hence, more advanced and robust turbulence models are needed in evaluating the turbine blade performance.

Most CFD studies of VAWTs have utilized the two-equation turbulence models such as the $k-\varepsilon$ model and its variants. In particular, the realizable $k-\varepsilon$ model (which compared to standard $k-\varepsilon$ model, the constants in turbulence viscosity formulation, C_μ , is not constant but a variable and the dissipation rate, ε , is derived from an exact equation for the transport of the mean-square vorticity fluctuation) is commonly used as it can produce reasonable good results for swirling flows, rotating and separating flows, boundary layers under strong adverse pressure gradients, and separated and recirculated flows, compared to the standard $k-\varepsilon$ model (Castelli, et al., 2010; Castelli, et al., 2011; Trivellato & Castelli, 2014; Mohamed, et al., 2015). For example, the results by (Castelli, et al., 2011) demonstrated that the realizable $k-\varepsilon$ model can predict the **power coefficient** (C_p) and **the optimum ratio between the tangential speed of the tip of a blade and the actual speed of the incoming wind, Tip Speed Ratio (TSR)**, in good agreements with the experimental data, even though it overestimated the power coefficient in the lower TSR range by a factor of 2 compared to test data. The simulation of (Ferreira, et al., 2010) also showed that the standard $k-\varepsilon$ turbulence model was able to predict the time-averaged vertical velocity distributions and the roll-up of the trailing-edge vortex shedding at the right phase angle (at around 120° azimuthal position), compared to that produced by the Spalart-Allmaras (SA) turbulence model.

Another two-equation turbulence model often utilized in VAWTs simulation is the $k-\omega$ shear stress transport ($k-\omega$ SST) model. For example, the study by (Wang, et al., 2018) showed that the $k-\omega$ SST model could produce the C_p curve in alignment with the experimental results, by improving the prediction errors about 50% at low TSR and about 35% at high TSR ranges respectively, compared to the realizable $k-\varepsilon$ model. Similar results also obtained by other researchers using the $k-\omega$ SST turbulence model (Lam & Peng, 2016; Almohammadi, et al., 2015; Arab, et al., 2017). In addition, a few studies have applied Transition SST turbulence model (Lanzafame, et al., 2014; Bangga, et al., 2017a; Rezaeiha, et al., 2018). Compared to two-equation models, the prediction accuracy of Transition SST turbulence model was generally improved. While the predicted power coefficients were close to the experimental data in the low TSR range, it still overestimated the C_p value in the high TSR range. The cause of this discrepancy has not been fully understood yet.

To further improve the prediction accuracy, more advanced turbulence models, such as **Large Eddy Simulation** (LES), have to be used in VAWTs simulation. The LES model is based on spatially filtered equations thus it is accurate in time. It explicitly calculates the large-scale eddies which contain the majority of energy spectrum, which affects the main flow. For small-scale eddies, their effects on the flow are considered using a **Sub-grid Scale** (SGS) model due to the universal behaviour of turbulence (i.e. Kolmogorov hypothesis). This feature makes the LES model more suitable to predict the behaviour of the vortices associated with the large flow separation and the dynamic stall in VAWTs. Despite of its capability for better prediction of flow characteristics around rotating bodies such as VAWTs, LES study for VAWTs applications are still very limited, mainly due to

its expensive computational cost (Li, et al., 2013; Ghasemian & Nejat, 2015; Elkhoury, et al., 2015; Posa & Balaras, 2018). To overcome this difficulty, a hybrid RANS-LES model like **Detached Eddy Simulation** (DES), **Improved Delayed Detached Eddy Simulation** (IDDES), and **Wall-Modelled LES** (WM-LES) are developed and utilized by many researchers in VAWTs applications (Lam & Peng, 2016; Lei, et al., 2017; Peng & Lam, 2016). These models still utilize RANS turbulence model in the near wall region to model small eddies, while switching to LES to accurately simulate large eddies in the intermediate and the far flow fields, including separated shear layer and wake regions (Liu, et al., 2017).

Recently, a **Stress-blended Eddy Simulation** (SBES) turbulence model has been developed by (Menter, 2018). This is an improved turbulence model based on DES and/or IDDES concept, to address some numerical issues observed in previous DES such as **Grid-induced Separation** (GIS). The latter is mainly due to the tendency of ‘shielding’ the boundary layer to be solved with the RANS mode and slowly “transition” from the RANS zone to the LES zone in **Separating Shear Layers** (SSLs) (Frank & Menter, 2017). Instead, the SBES model uses an improved ‘shielding’ function to protect the RANS boundary layers and to switch to an existing algebraic LES model in the LES zone. As a result, the RANS and the LES zones can be clearly distinguished by visualizing the ‘shielding’ function. Moreover, due to the lower turbulence stress level enforced by the LES model, the SBES model can reduce the transition time from the RANS to the LES in SSLs, which can produce better, realistic, and consistent solutions. Furthermore, this turbulence model allows a RANS-LES “transition” even on a coarser grid that other DES models cannot.

To authors’ knowledge, it is not yet found in public domain about the application of SBES turbulence model for CFD simulation of VAWTs. Nevertheless, this turbulence model has been successfully applied in the CFD simulations of rotating devices (Ravelli & Barigozzi, 2018a; Ravelli & Barigozzi, 2018b; Cai, et al., 2019). They reported that SBES model can produce the better predictions compared to other hybrid RANS-LES models (Ravelli & Barigozzi, 2018b). Furthermore, SBES model can generate finer turbulence structures with abundant vortex structures. In comparison with other hybrid RANS-LES models, SBES model can capture faster development of turbulence and more ordered turbulence structures (Cai, et al., 2019). Hence, this paper will be the first attempt of this kind to simulate a three-straight-bladed VAWT using the SBES turbulence model in order to access its capability of producing accurate CFD prediction and to evaluate its sensitivity on grid topology change.

2. Methodology

2.1 The **Vertical Axis Wind Turbine (VAWT)** model

Two-dimensional (2D) simulations of a three-straight-bladed VAWT featured in a study by (Castelli, et al., 2011) (see Figure 1) are carried out. **2D simulation is chosen to decrease the computational cost as this study only limited to evaluate the ability of SBES turbulence model to predict flow around VAWT. Even though three-dimensional (3D) simulation is preferable for flow with separation, previous study using hybrid LES-RANS model (DES) showed that 2D simulation could generate similar results with 3D simulation in turbulent separation case (Travin, et al., 2000). Hence, it is reasonable to perform 2D simulation by using SBES turbulence model as the model is actually based on DES model.**

The blades use NACA 0021 aerofoil as cross-section profile. Three turbine blades rotate in counter-clockwise direction around a supporting rod in the centre. The pressure centre of the blade is set at 0.25 chord from the leading edge of the aerofoil (**The place where the rod is connected to the blade**). The geometry details are summarized in Table 1. The rotor blade azimuthal position is calculated based on the angular coordinate from the pressure centre of blade 1. In the experiments performed by (Castelli, et al., 2011), the turbine was operated at several angular velocities (ω) with a constant incoming wind speed (U_∞) of 9 m/s. The angular velocity ω is used to define *TSR* as,

$$TSR = \frac{\omega R}{U_\infty} , \quad (1)$$

where R is the turbine radius, i.e. from the blade pressure centre to the supporting rod centre.

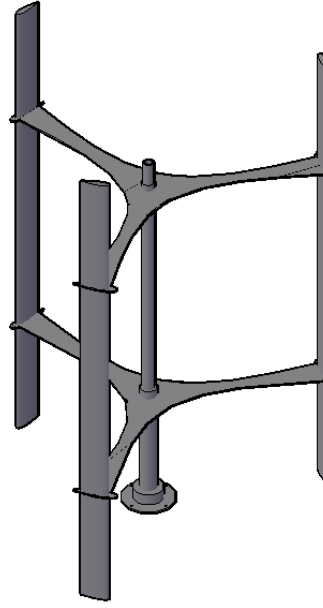


Figure 1. Experimental VAWT model (all measurements can be seen in Table 1) based on (Castelli, et al., 2011).

2.2 Turbulence modelling

This CFD investigation uses SBES turbulence model, which revises the shielding function of the shielded DES (SDES) SST model to switch between the LES and the RANS zones automatically. While the blending function remains the same as that of the shielding function SDES (f_{SDES}), in the LES zone where $f_{SDES} = 0$, SBES introduces an explicit model switching to an algebraic LES. By adding equation (2) shown below, the SBES model achieves a smooth blending of the Reynolds stress between the RANS and the LES formulations (Frank & Menter, 2017) as:

$$\tau_{ij}^{SBES} = f_{SDES} \tau_{ij}^{RANS} + (1 - f_{SDES}) \tau_{ij}^{LES}, \quad (2)$$

where τ_{ij}^{RANS} is the RANS Reynolds stress tensor and τ_{ij}^{LES} is the LES stress tensor. If the model is based on the eddy viscosity concept, this equation can be further simplified as:

$$\nu_t^{SBES} = f_{SDES} \nu_t^{RANS} + (1 - f_{SDES}) \nu_t^{LES}. \quad (3)$$

For the RANS model, the transition SST turbulence model will be used.

Table 1. Main geometrical features of experiment model (Castelli, et al., 2011).

Rotor diameter (D_{rotor} (mm))	1030
Rotor height (H_{rotor} (mm))	1456.4
Rotor swept area (A_s (m ²))	1.236
Number of blade (N (-))	3
Blade profile	NACA 0021
Chord length (c (mm))	85.8
Spoke-blade connection	0.5c
Solidity (σ (-))	0.5
Wind tunnel height (H_{wind} (mm))	4000
Wind tunnel wide (W_{wind} (mm))	8000

Table 2. Simulation setup

Parameters	Current Simulation
Fluid	Incompressible
Model	2D
Solver	Pressure-based; Unsteady
Models	Transitional SST SBES
Methods	Pressure-Velocity Coupling
	Scheme : SIMPLE
	Spatial Discretization
	Gradient : Least Squares Cell Based
	Pressure : Standard
	Momentum : Bounded Central Differencing
	Turbulence Kinetic Energy: Second Order Upwind
	Specific Dissipation Rate: Second Order Upwind
	Intermittency: Second Order Upwind
	Momentum Thickness Re: Second Order Upwind
	Transient Formulation : Second Order Upwind
Residual convergence criterion	10^{-6}
Initialization	Hybrid Initialization
Iteration every time step	40
Time Step	1°
Number of revolutions	34

2.3 Simulation setup

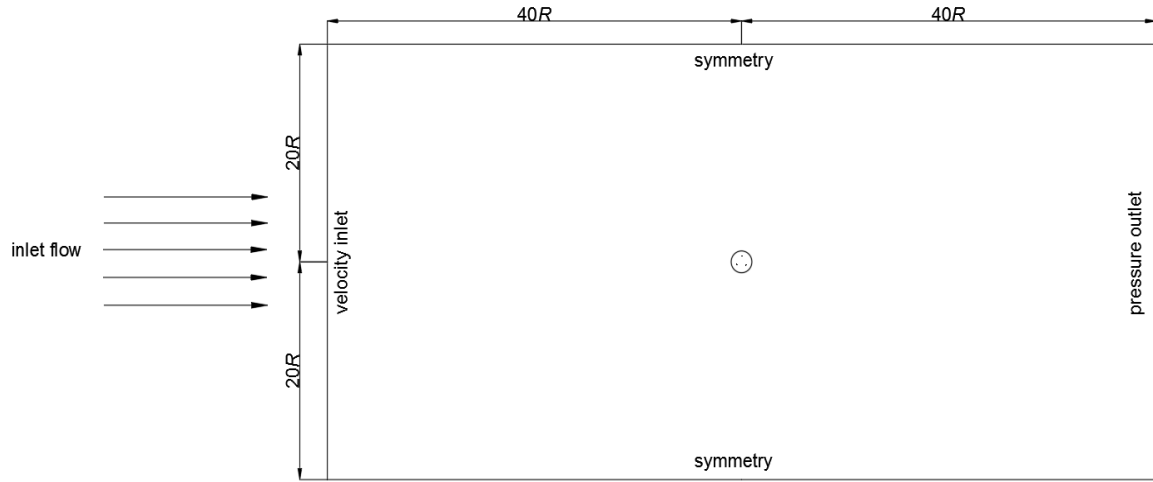
The simulation runs in unsteady mode using a pressure-based solver. Table 2 provides the details of solution methods and settings. All residuals are set to be 10^{-6} . According to the numerical study by (Castelli, et al., 2011), the rotor height and the rotor swept area are adjusted to be 1000 mm and 1.03 m^2 respectively for 2D modelling, and the time step is set to equal 1° rotation of the rotor blade. The total simulation time is determined by the rotor revolution, which should be sufficiently long in order to allow the wake unsteadiness and the periodic motion of the rotating blades to be fully developed.

Table 3. Grid discretization

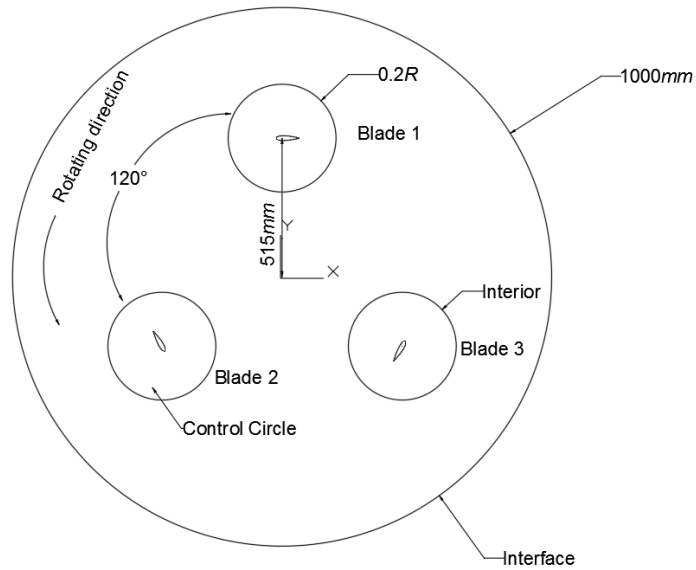
	O-grid	C-grid
Type of Shape		
Far field	Rectangular	C combined with rectangular
Rotating Core	Circle	Circle
Control Circle	Circle	C combined with rectangular
Type of Grid		
Far field	Quadrilateral structured grid	Quadrilateral structured grid
Rotating Core	Quadrilateral dominant grid	Quadrilateral dominant grid
Control Circle	Quadrilateral structured grid	Quadrilateral structured grid
Total number of cells		
Far field	34200	18240
Rotating Core	22527	22527
Control Circle	20880	49680
Growth Rate	1.2	1.2
Element around body	174	174
Element around trailing edge	14	14
Body sizing for rotating core	12 mm	12 mm

2.4 Computational domain and grid discretization

The study starts with grid sensitivity analysis with two types of grid (denoted as O-grid (OG) and C-grid (CG) thereafter). Table 3 lists the details of the grid discretization for these two grids. The aim is to identify an appropriate grid topology and resolution for the main VAWT simulation using SBES turbulence model. Figures 2 and 3 show the computational domain which consists of three sub-domains, namely far field, rotating core and control sub-domains for OG and CG meshes, respectively. The specifications of the domain and the grid generation are described below.

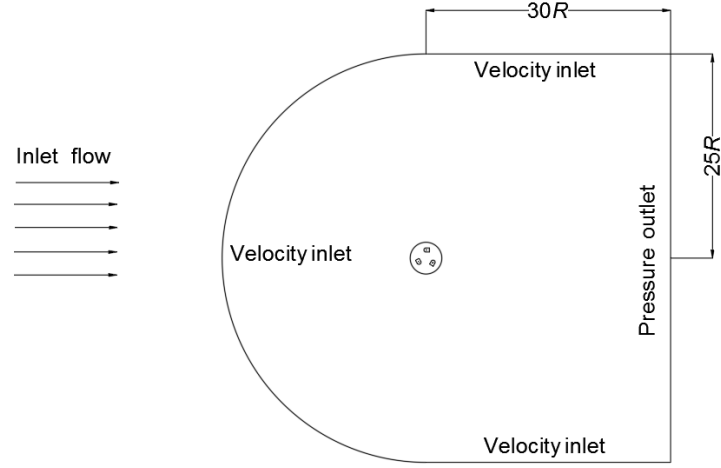


(a) Overview of the computational domain

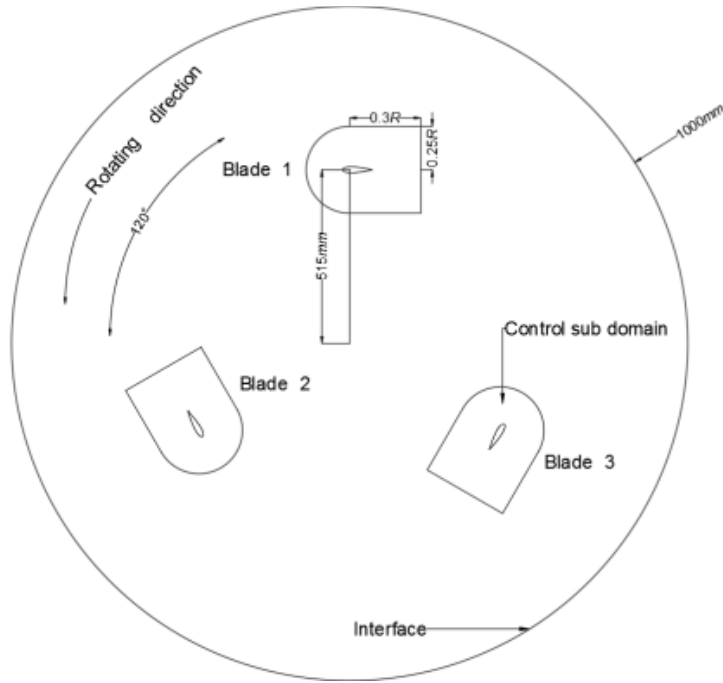


(b) Rotating core sub-domain

Figure 2. Detailed computational domain and sub-domains of O-grid.



(a) Overview of the computational domain



(b) Rotating core sub-domain

Figure 3. Detailed computational domain and sub-domains of C-grid.

2.4.1 Far field sub-domain

This is non-rotating sub-domain surrounding the rotating core sub-domain. It has different shapes for each type of grid as explained below.

O-grid

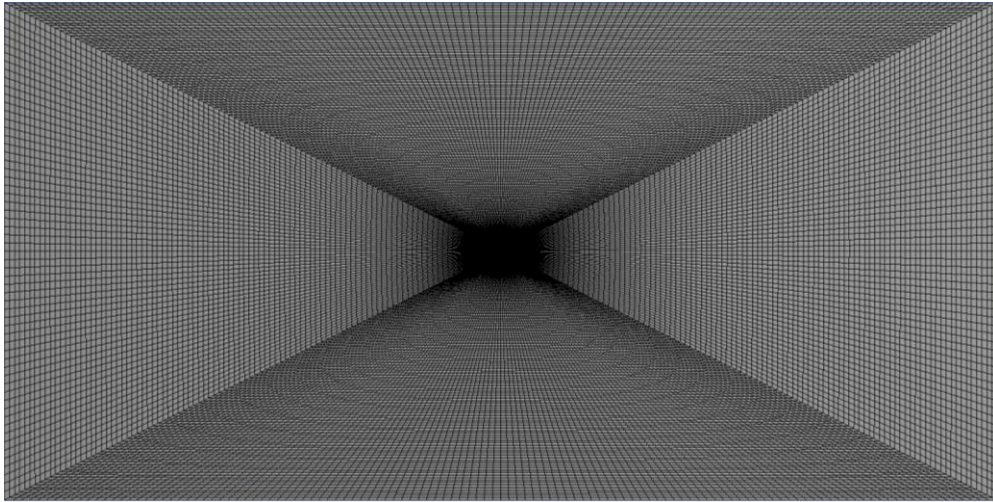
The O-grid uses rectangular far field sub-domain as suggested in previous studies (Castelli, et al., 2011; Wang, et al., 2018; Lam & Peng, 2016). Based on (Wang, et al., 2018), to avoid the boundary condition influences, the inlet and the outlet boundaries are located at $40R$ away from the centre of the domain, while the side walls are located at $20R$ away from the centre of turbine rotating axis, respectively. The velocity inlet and the pressure outlet boundary conditions are applied. Meanwhile, the side walls are defined as symmetric boundaries. A structured grid with quadrilateral cells is generated in this sub-domain (see Figure 4a).

C-grid

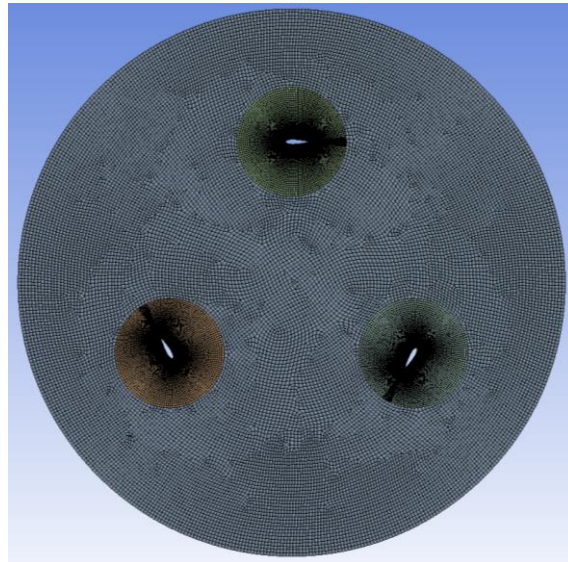
The C shape together with a rectangular enclosure is used for the far field sub-domain. The C shape has $25R$ in radius and the rectangular enclosure has $30R$ in stream wise distance from the centre of the turbine rotating axis to the exit, as suggested by (Zhu, et al., 2018). Figure 5a has shown this sub-domain is divided into 6 regions to facilitate smooth grid discretization. Similar to the O-grid, a structured grid with quadrilateral cells is generated within this sub-domain (see Figure 5b).

2.4.2 Rotating core sub-domain

This sub-domain is fluid region and utilised to implement the revolution of the rotor blade. It has 2000 mm in diameter and rotates in anti-clockwise direction around the turbine rotating axis at a given angular velocity. To ensure the continuity of fluid flow in the far field and the rotating core sub-domains, a ‘fluid-fluid’ interface is set up at the boundary intersection of these two sub-domains. The two types of grid both utilise quadrilateral dominant elements (see Figures 4b and 5c).

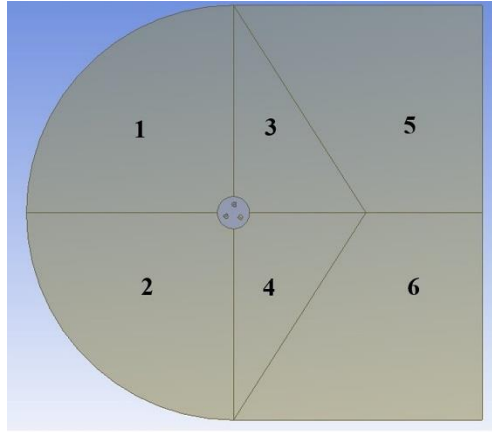


(a) Far field

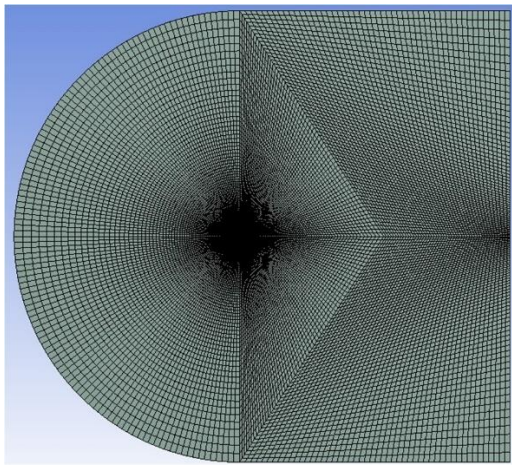


(b) Rotating core

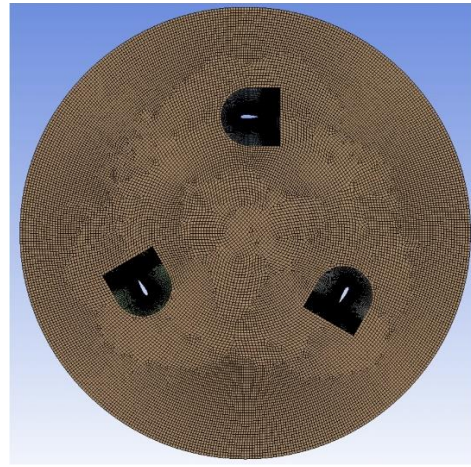
Figure 4. Grid details in two sub-domains of O-grid.



(a) Partition of farfield sub-domain



(b) Far field sub-domain



(c) Control sub-domain

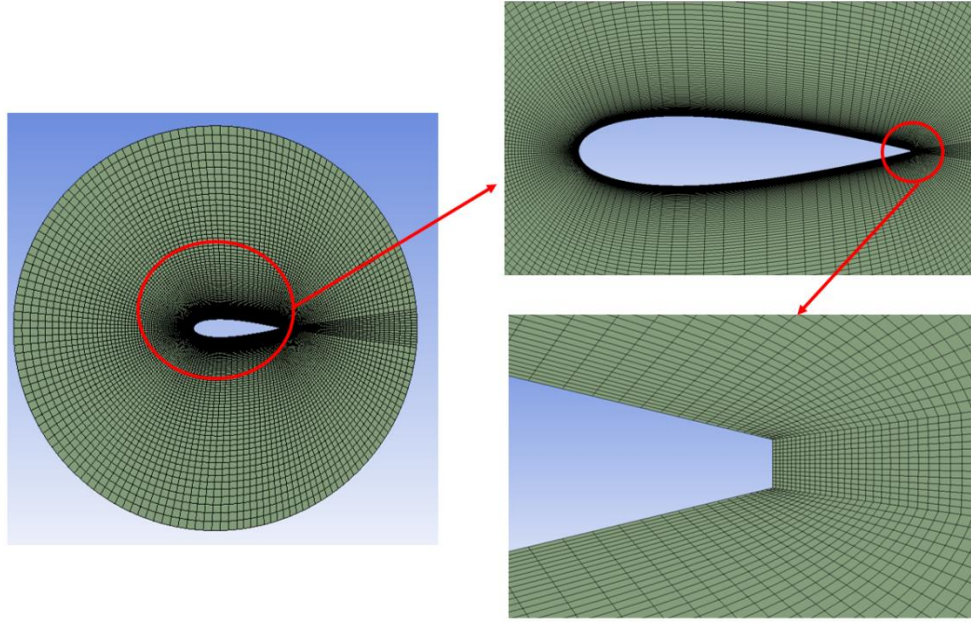
Figure 5. Grid details in two sub-domains of C-grid.

2.4.3 Control sub-domain

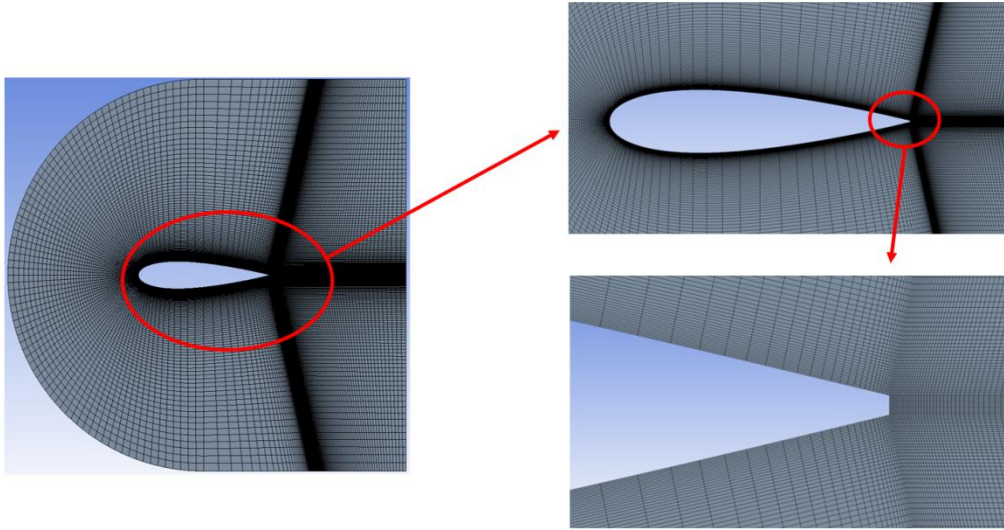
This sub-domain is used to generate meshes around the blades. Three control domains with inserted blades are located inside the rotating core and separated by 120° angular distance between the adjacent blades. The boundary is also interpreted as “interior” to ensure the continuity of the fluid flow.

For OG, each control sub-domain has a circle shape with a radius of 200 mm , in which a structured O-grid around a blade is generated. For CG, the C-shape has $0.25R$ in radius and $0.3R$ in length from the centre of the blade. It uses the C-grid around the blade with gradually increased grid size.

The structured quadrilateral cells are generated in this sub-domain, with fine grids in the near wall region (see Figures 6a and 6b) and coarse grids away from the wall. When transition SST turbulence model is used, it is necessary to generate the first layer height to satisfy the criteria of non-dimensional wall distance $\Delta y_l^+ < 1$.



(a) O-grid



(b) C-grid

Figure 6. Grid around blade wall.

3 Results and Discussion

3.1 Revolution convergence

In VAWT simulation, it is necessary to collect data samples after reaching a statistically converged flow field. This can be done by monitoring time history of the moment coefficient (C_m) or the power coefficient (C_p) defined as:

$$C_m = \frac{M}{\frac{1}{2}\rho U_\infty^2 A_s R}, \quad (4)$$

$$C_p = TSR \times C_m, \quad (5)$$

where M (N) is the predicted moment force, ρ (kg/m^3) is the fluid density, A_s (m^2) is the rotor swept area.

In previous URANS simulation, (Castelli, et al., 2011) started the data sampling while the C_m variations between two neighbouring revolutions is less than 1%. Another study by (Rezaeiha, et al., 2018) found that after 20 revolutions the changes of C_m and C_p between two successive revolutions could be below 0.1% and 0.2% respectively, and between 20 and 100 revolutions, the differences over C_m and C_p would be lower than 1.06% and 2.41%, respectively. While in agreement with those observations, the present study also found that after initial 20 revolutions, the C_m variation reduced to less than 0.45% compared to previous revolution, indicating that a good convergence has been achieved.

However, due to the differences between the URANS and the SBES turbulence models, further test is needed to verify the revolution convergence of the SBES models. As shown in Figure 7, simulations have achieved convergence status after 34 revolutions for both OG and CG meshes. After this revolution, the difference of average power coefficient between two neighbouring revolutions is only 0.001%. Hence, for all the remaining simulations, data retrieval will be collected from the 35th revolution.

Compared to the URANS, SBES turbulence model will take more revolutions to reach convergence status. It is probably due to the fact that URANS turbulence models are mainly solving the mean flow and those large flow motions in the near field, and use ensemble averaging solution in the far field (Salim, et al., 2013). In contrary, SBES turbulence model utilises the LES model in the far field, which can resolve the flow fluctuations to some extents and as a result, it will take longer time to achieve statistically converged flow field for both near and far fields.

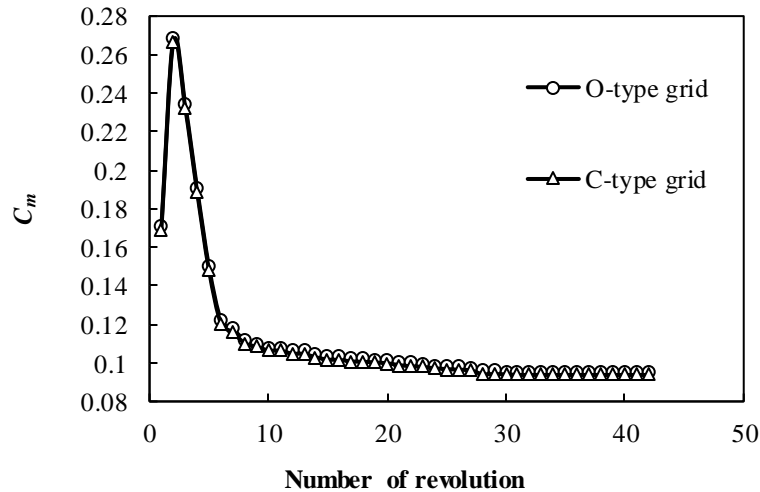


Figure 7. Moment coefficient changes over turbine revolution.

3.2 Grid convergence

The grid convergence study is conducted for simulation at $TSR = 3.09$. At first, O-grid is considered with three grid resolutions from coarse, medium to finer meshes, each having 87, 174 and 348 cells around the blade. Then, C-grid is also tested using three grid resolutions with the same number of cells around the blade as the O-grid.

Figures 8 and 9 illustrate the comparison of instantaneous moment coefficients over one revolution for both OG and CG meshes, respectively. As shown in the both figures, the moment coefficient changes along a azimuthal positions have shown little difference between the medium and the fine grids while the coarse grid could not produce satisfying instantaneous moment coefficients. For both OG and CG, the average power coefficients of medium and fine grids are in good agreement with the experimental results of (Castelli, et al., 2011). Moreover, the relative error of average power coefficients between the medium and the fine grids is less than 4%. Therefore, the medium grid has been chosen for the rest of simulations.

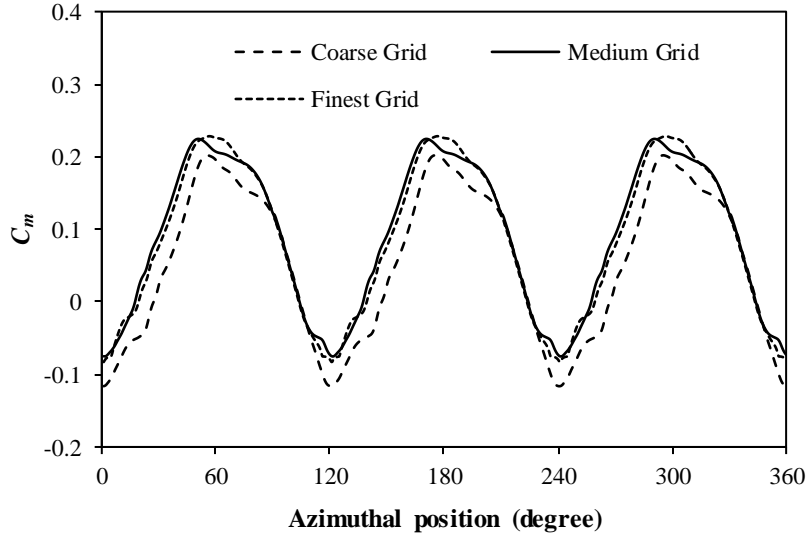


Figure 8. Comparison of instantaneous moment coefficients of VAWT with different grid resolutions for O-grid.

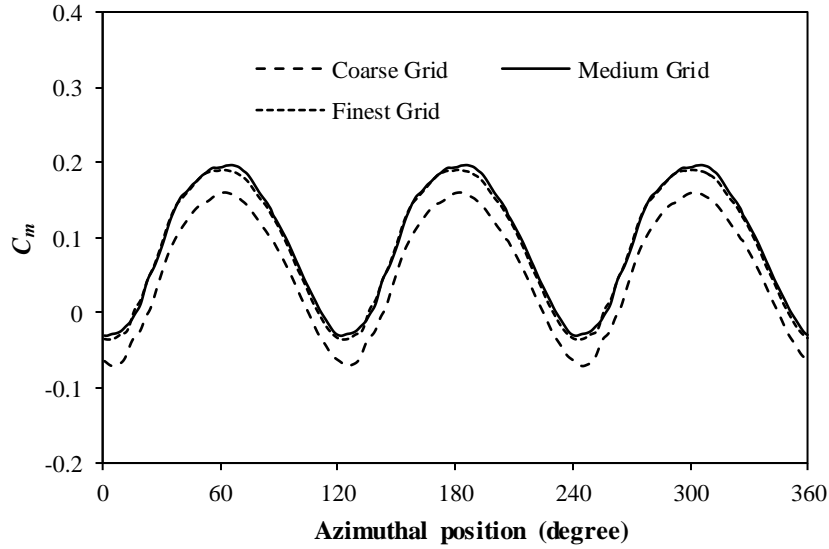


Figure 9. Comparison of instantaneous moment coefficients of VAWT with different grid resolutions for C-grid.

3.3 Results validation

For validating SBES results, simulations are performed for TSR ranging from 1.44 to 3.3. Figure 10 shows the average C_p prediction of current CFD predictions, along with experimental data and CFD results of (Castelli, et al., 2011). Compared to CFD results of (Castelli, et al., 2011), the current CFD results using the same turbulence model of (Castelli, et al., 2011) (i.e. URANS realizable $k-\varepsilon$ with enhanced wall treatment) give better prediction especially in high TSR . However, these results still give large errors compared to experiment results.

On the other hand, it is found that present CFD prediction with OG and CG topologies using SBES turbulence model successfully reproduces the C_p curve, especially the maximum peak value at an optimum TSR (2.64), is in very good agreement with the experiment of (Castelli, et al., 2011). It is clear that CFD using SBES turbulence model gives much better C_p predictions than using URANS realizable $k-\varepsilon$ turbulence model with enhanced wall treatment.

Figures 11 and 12 illustrate the results comparison of the average power coefficient and the relative error between experiments and CFD results of (Castelli, et al., 2011) and present study over one revolution. As shown in Figure 11, even though the current CFD model can decrease the prediction error, the realizable $k-\varepsilon$ turbulence model with enhanced wall treatment still gives considerable error, especially at low TSR range. In fact, the angle

between the blade zero lift line and the freestream direction (defined as absolute value of angle of attack, AoA) are relatively larger at low TSR than high TSR (Ma, et al., 2018). Moreover, the blades of VAWT can experience large range of AoA at the same time. In addition, flow around blades could experience enormous large viscous region at low $TSRs$ due to low Reynolds number effects (Lei, et al., 2017). Therefore, the blades will experience deep dynamic stall under such conditions. Because the $k-\epsilon$ turbulence models family will over-predict the turbulence kinetic energy, it is unable to predict the deep dynamic stall at high AoA (Lei, 2005). As a result, URANS turbulence model produces large error at low $TSRs$.

Compared to the URANS, SBES turbulence model produces relatively smaller error in all range of $TSRs$ for both O-grid and C-grid (see Figure 12). It is due to the fact that the large eddies in the far field are resolved by the LES model in the SBES, while in URANS, the turbulence is treated as isotropic, leading to the momentum transport in the far field cannot be correctly considered. It is mentioned by (Warhaft, 2000) that this isotropic treatment can lead to reveal a greater degree of intermittency (As example, URANS simplification make turbulent heat fluxes have basically no effect on the mean temperature. However, in fact, the turbulent fluctuations may generate significant difference properties in space and instantaneous properties in time (McDonough, 2007)). Furthermore, as discussed by (Lei, et al., 2017), DES turbulence model family can perform better due to its ability to produce realistic average wake velocity in the near and far fields.

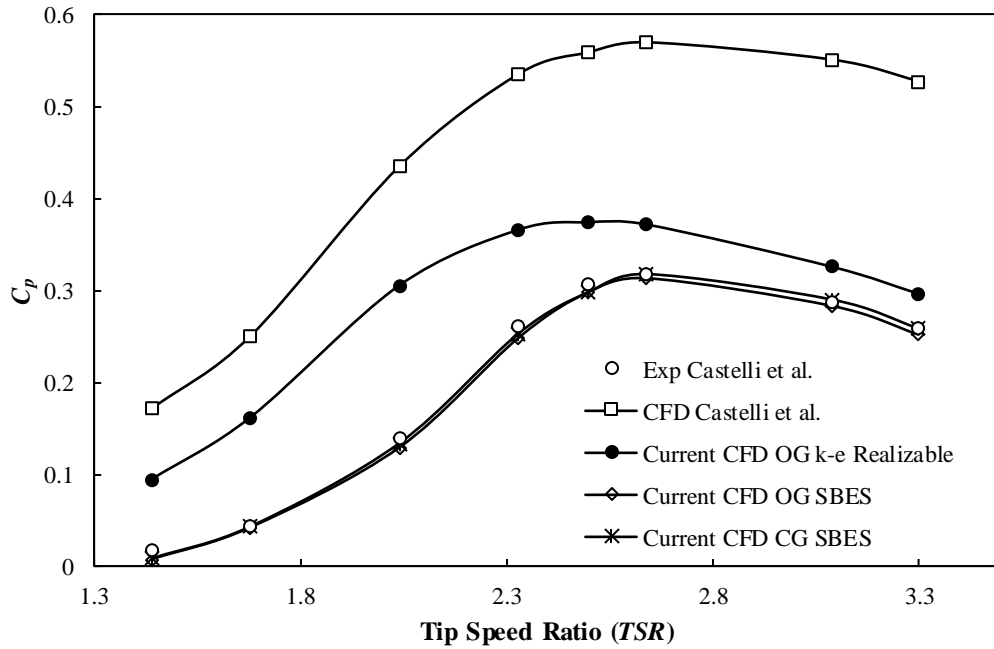


Figure 10. C_p prediction comparison of current CFD and (Castelli, et al., 2011) models.

To understanding better ability of SBES turbulence models to predict C_p of VAWT, the contour plots of z-vorticity at several important azimuthal positions of blade 1 is illustrated in Figure 11. It is shown that SBES turbulence model can predict stronger vortex shedding compared to $k-\epsilon$ realizable turbulence model. It is also noticeable that compared to $k-\epsilon$ realizable, SBES can show the development of dynamic stall and roll up trailing edge vortices. This ability lead to lower prediction of instantaneous C_m value almost in all azimuthal position and stronger fluctuation of this value after vortex shedding region therefore resulting lower prediction of C_p (see Figure 12).

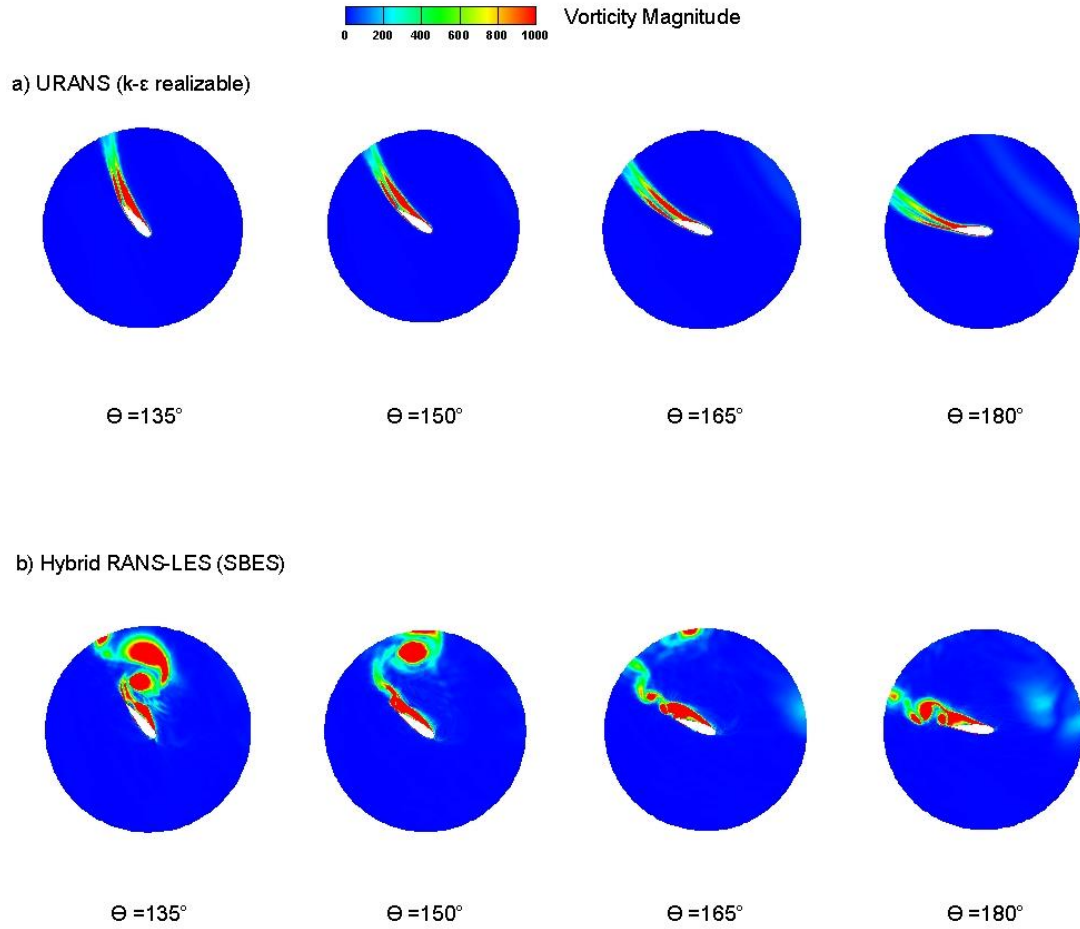


Figure 11. Comparison of contourplots of z -vorticity, indicating the process of flow separation at important azimuthal positions, prediction between SBES and k - ϵ realizable turbulence models.

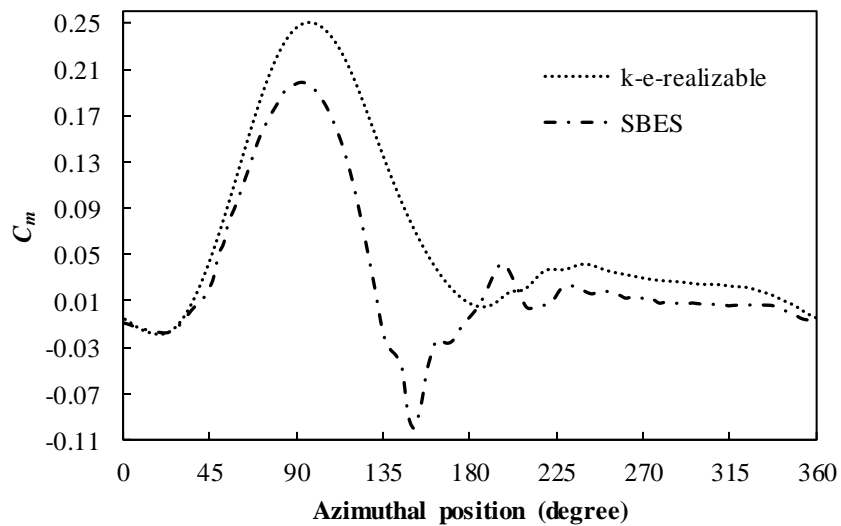


Figure 12. Comparison of instantaneous moment coefficient distribution of SBES and k - ϵ realizable turbulence models.

3.4 Grid topology study

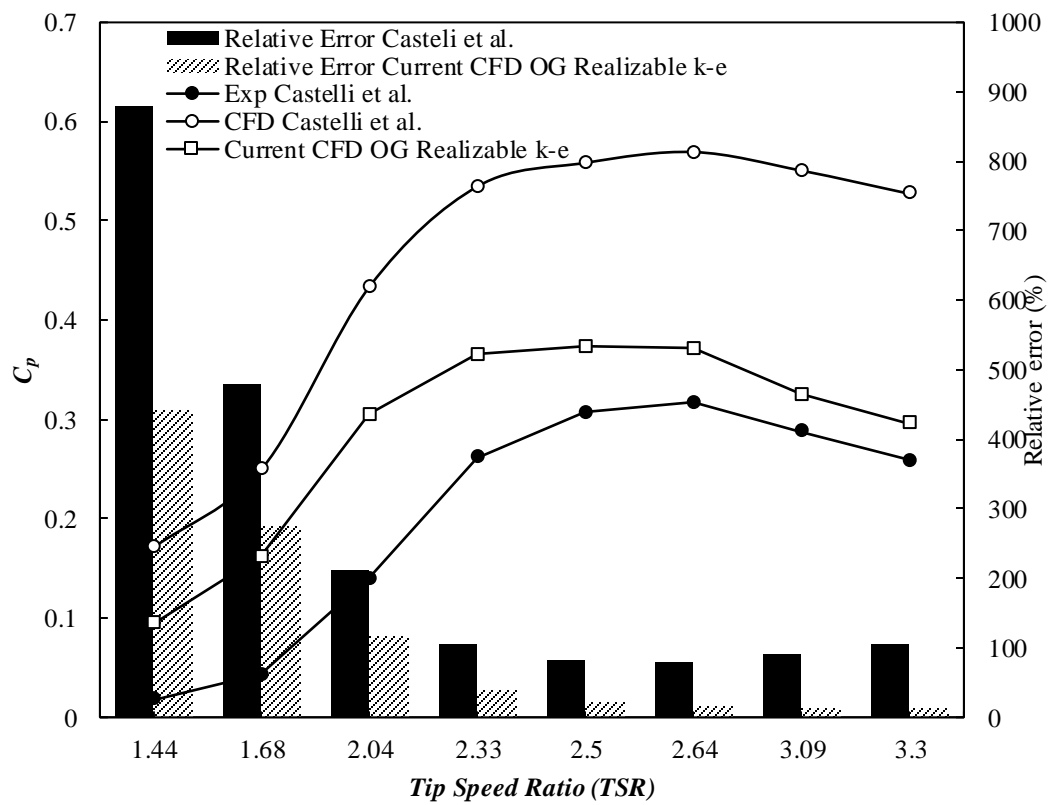


Figure 13. Comparison of average power coefficients between the experiment and simulation of (Castelli, et al., 2011) as well as relative errors in percentage.

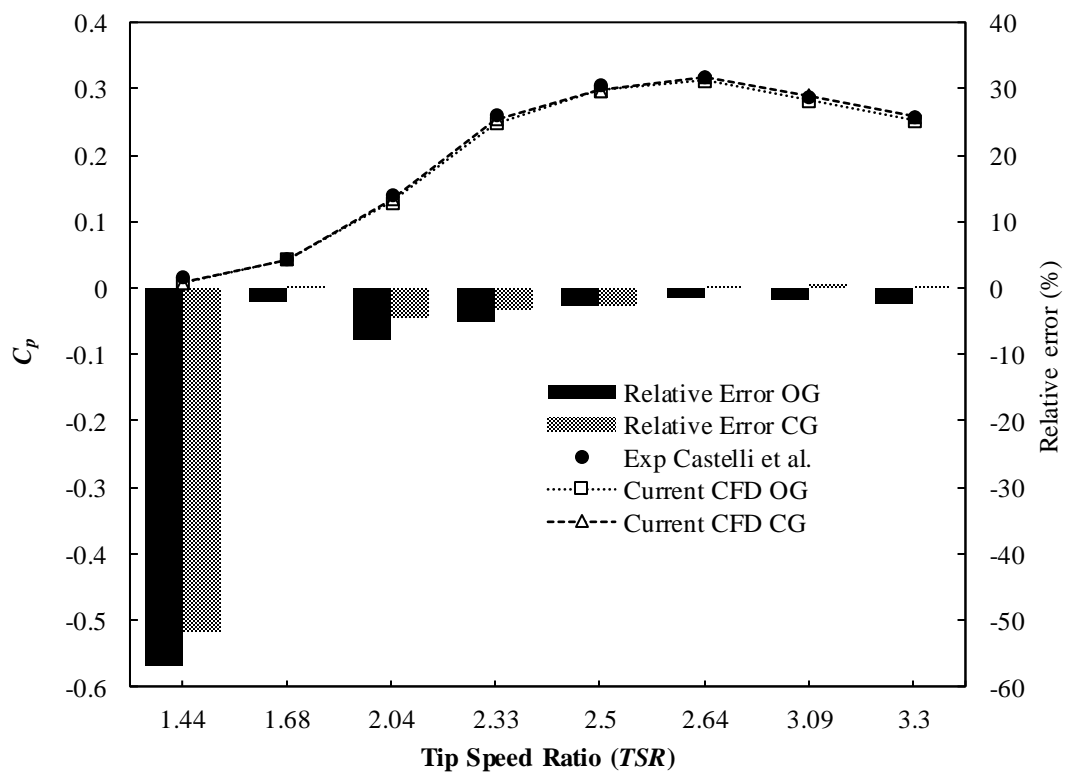


Figure 14. Comparison of average power coefficients between the experiment of (Castelli, et al., 2011) and current CFD simulation as well as relative errors in percentage.

As depicted in Figures 13 and 14, there are clear differences in the time average C_p distribution between CG and OG meshes, despite that the tendency of general behaviour of C_p distribution is predicted consistently by the two grid topologies. Overall, the discrepancy between the OG and CG topologies is relatively minor if the time step is small enough as illustrated in Figures 13 and 14 respectively.

For all range of $TSRs$, simulations using the CG mesh produces relatively smaller error than that of OG, compared to experiment. This is probably contributed by the effect of grid density in the close region of the blade (i.e. control sub-domain) rather than the grid topology. Note that the CG control sub-domain contains more cells than OG (see Table 3) even though they have same number of cells around the blade. Furthermore, DES turbulence model family (including SBES) is relatively sensitive to grid resolutions not only in near wall but also in far field. As a result, simulation of CG mesh gives better prediction than OG. This observation is in agreement with a previous study of HAWT blade using DDES (Bangga, et al., 2017b).

Instantaneous moment coefficients of one representative blade (i.e. blade 1) over one revolution are plotted in Figure 15 for OG and CG meshes at $TSR = 3.09$. Simulation of OG predicts earlier separation than CG, exhibited by earlier drop of C_m value below zero at around 130° azimuthal position, while simulation of CG starts to have negative C_m value later at about 150° azimuthal position (see a dashed circle in Figure 15). This means that the starting point of no torque production (i.e. no power generation) predicted by simulation using OG mesh is earlier than that of CG mesh. However, the predicted recovery points (i.e. starting to produce positive torque again) are similar (around 190° azimuthal position) and their behaviours after that point are almost identical. In addition, both simulations produce almost same maximum C_m values at same azimuthal position. Due to these differences in prediction, the predicted power generation with OG mesh is slightly lower than CG and experimental measurements.

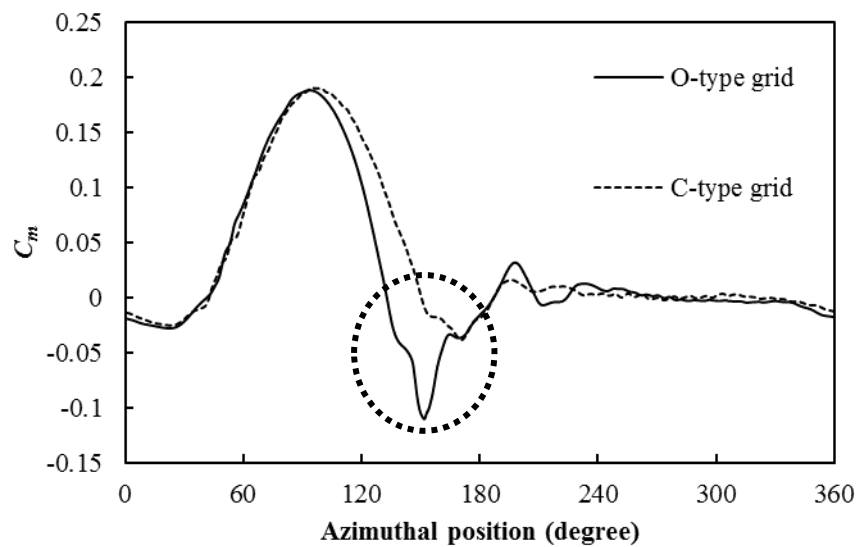


Figure 15. Comparison of instantaneous moment coefficient distribution of blade 1 for one turbine revolution between O-grid and C-grid.

4. Conclusions

Two-dimensional CFD studies of **Stress-blended Eddy Simulation (SBES)** turbulence modelling to predict the performance of a three-straight-bladed VAWT and to evaluate grid topology effect on prediction accuracy have been carried out for *Tip Speed Ratios (TSRs)* ranging from 1.44 to 3.3 at a constant wind speed 9 m/s. The simulations are performed for 34 rotor revolutions to reach statistically converged results. After that, additional one revolution is simulated to collect data samples for analysis.

The results have shown that for both **O-grid (OG)** and **C-grid (CG)**, SBES turbulence model can produce the power coefficients in agreement with experimental data (Castelli, et al., 2011). It is because SBES model resolves turbulence flow in the far field region with the time accurate **Large Eddy Simulation (LES)** model. Moreover, simulation with CG mesh produces better **power coefficient (C_p)** distribution than OG due to higher grid density of CG in the near blade region. It is also found that an earlier flow separation has been predicted by simulation

with OG mesh (at 130° azimuthal position) than CG (at 150° azimuthal position). However, both simulations using these two grids also predicted similar recovery points (i.e. around 190° azimuthal position). Hence, it produces relatively lower C_p than CG and experimental data (Castelli, et al., 2011). Further development of the present study will be carried out to perform 3D simulation in order to investigate the effect of 3D modelling contributing to the accuracy of CFD prediction using SBES model.

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